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The following are solutions to the problems found in Chapter 1 of John D. Anderson's 2004 book Modern Compressible Flow. At the nose of a missile in flight, the pressure and temperature are 5.6 atm and 850 ° R, respectively. Calculate the density and specific volume.

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From the differential form of the continuity equation we can write that, $\frac{d}{dt}(\rho u) + \frac{d}{dx}(\rho u^2) = 0$. From Euler's equation, $\rho \frac{dp}{dx} = -\rho u \frac{du}{dx}$. We are considering an adiabatic, inviscid flow (isentropic flow).

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Anderson's book provides the most accessible approach to compressible flow for Mechanical and Aerospace Engineering students and professionals. In keeping with previous versions, the 3rd edition uses numerous historical vignettes that show the evolution of the field.

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